SECTION II - PERFORMANCE

For various reasons it has not been possible to carry out performance tests by B.A.C. pilots, the chief of these being that the only British aircraft available has unpressurised magnetos, priority for this modification having been given to the Service Squadrons. This precludes operation at the altitudes required for performance measurement.

However, the U.S. Navy Flight Test Section, at Patuxent River, have carried out extensive performance tests and their report is reproduced below, together with performance curves.

The U.S. Army also carried out brief performance tests at Wright Field, and a copy of their report is reproduced, following that by the U.S. Navy. Where applicable, the points obtained by the U.S. Army have been superimposed on the U.S. Navy curves. These show extremely good agreement.

"Subj: Model F4U-1 Airplane - Performance Characteristics of.

1. This memorandum with its attendant enclosure is submitted in compliance with the request of the British Air Commission for information on the performance characteristics of the Model F4U-1 airplane.

2. Enclosure 1 consists of a series of performance curves on the model F4U-1 airplane No. 02155, when loaded as a normal fighter at a gross weight of 11,194 pounds. The series contains the following curves:

(a) Time to climb and rate of climb, at normal and at military* rated powers, plotted against standard altitude.

(b) Indicated brake horsepower required, at normal and at military* rated powers, plotted against true indicated airspeed.

(c) Indicated airspeed plotted against true indicated airspeed.

(d) Maximum speed, at normal and at military rated powers, plotted against standard altitude.

(e) Brake horsepower available, at normal and at military rated powers, plotted against standard altitude.

(f) Rate of climb at altitude as a function of cowl flap opening plotted against indicated airspeed.

* See table on page 19 for details of engine ratings.

Courtesy Neil Stirling
3. A summary of the performance characteristics of the Model F4U-1 airplane No. 02155, at a gross weight of 11,194 pounds, is presented as follows:

(a) Climb characteristics using best climbing speed, minimum cowl flap opening, and normal rated power:

1. Rate of climb at sea level...... 2,200 fpm

2. Rate of climb at airplane critical altitude in neutral blower (5,500 ft.) ............... 2,160 fpm

3. Rate of climb at airplane critical altitude in low blower (17,000 ft.) ............... 1,980 fpm

4. Rate of climb at airplane critical altitude in high blower (21,200 ft.) ............... 1,760 fpm

5. Service ceiling (rate of climb = 100 fpm) ............... 38,000 ft.

(b) Climb characteristics using best climbing speed, minimum cowl flap opening, and military rated power;

1. Rate of climb at sea level ...... 2,890 fpm

2. Rate of climb at airplane critical altitude in neutral blower (700 ft.) ............... 2,880 fpm

3. Rate of climb at airplane critical altitude in low blower (15,400 ft.) ............... 2,300 fpm

4. Rate of climb at airplane critical altitude in high blower (21,200 ft.) ............... 1,840 fpm

5. Service ceiling (rate of climb = 100 fpm) ............... 38,200 ft.

(c) Maximum true airspeed using normal rated power;

1. At sea level ................. 326 mph

/ (2)

Courtesy Neil Stirling
(2) At airplane critical altitude in neutral blower (5,900 ft.) .... 343 mph

(3) At airplane critical altitude in low blower (19,700 ft.) .... 395 mph

(d) Maximum true airspeed using military rated power;

(1) At sea level ................. 348 mph

(2) At airplane critical altitude in neutral blower (1,400 ft.) .... 352 mph

(3) At airplane critical altitude in low blower (17,800 ft.) .... 390 mph

(4) At airplane critical altitude in high blower (22,800 ft.) .... 295 mph

(e) Rates of climb at 5,000 feet, using military power and cowl flaps open; - ONE THIRD, TWO THIRDS, FULL

(1) At IAS of 120 kts 2,360 fpm 2,310 fpm 2,240 fpm

(2) At IAS of 140 kts 2,280 fpm 2,190 fpm 2,130 fpm

(3) At IAS of 160 kts 2,120 fpm 2,060 fpm 1,900 fpm

(f) Rates of climb at 15,000 feet, using military power and cowl flaps open; - ONE THIRD, TWO THIRDS, FULL

(1) At IAS of 120 kts 2,300 fpm 2,220 fpm 2,010 fpm

(2) At IAS of 140 kts 2,200 fpm 2,090 fpm 1,960 fpm

(3) At IAS of 160 kts 1,990 fpm 1,900 fpm 1,640 fpm

(g) Rates of climb at 25,000 feet, using military power and cowl flaps open; - ONE THIRD, TWO THIRDS, FULL

(1) At IAS of 120 kts 1,480 fpm 1,360 fpm 1,160 fpm

(2) At IAS of 140 kts 1,340 fpm 1,120 fpm 990 fpm

(3) At IAS of 160 kts 1,120 fpm 760 fpm 630 fpm

4. All available performance characteristics of the Model F4U-1 airplane which are not presented in enclosure 1 are tabulated as follows:

(a) Stalling speeds, airplane No. 02156 at a gross weight of 11,499 pounds:

(1) No power, clean condition .......... 102,5 mph
Section II - Performance

(2) No power, landing condition ...... 85.0 mph
(b) Take-off data, airplane No. 02158 at a gross weight of 11,190 pounds;
   (1) True indicated airspeed at take-off 78.0 mph
   (2) Take-off distance, zero wind ...... 560 ft.
   (3) Take-off distance, 25 knot wind .... 240 ft.

Encl,
1. Series of curves entitled, "Model F4U-1 Airplane No. 02155, Performance Characteristics, Normal Fighter, Gross Weight = 11,194 Lbs." (3 sheets)

/s/ PAUL H. RAMSEY
Flight Test Officer
MODEL F4U-1 AIRPLANE NO 02155
PERFORMANCE CHARACTERISTICS
NORMAL FIGHTER
GROSS WEIGHT=111940LB

MAXIMUM SPEED AT
RATED POWER

PERFORMANCE CHARACTERISTICS
OF P&W R-2800-8
ENGINE AS INSTALLED
IN AIRPLANE

U.S. ARMY POINTS

STANDARD ALTITUDE - THOUSANDS OF FEET
0 2 4 6 8 10 12 14 16 18 20 22 24 26 28 30 32 34

MILITARY POWER
NORMAL POWER

TRUE AIRSPEED - MPH
320 330 340 350 360 370 380 390 400

BHP AVAILABLE - HP
800 1000 1200 1400 1600 1800 2000

Courtesy Neil Stirling
PRATT & WHITNEY R-2800-8 ENGINE

ENGINE RATINGS

The following table gives the ratings employed during the U.S. Navy Corsair performance tests:

<table>
<thead>
<tr>
<th>Blower</th>
<th>Rating</th>
<th>H.P.</th>
<th>R.P.M.</th>
<th>Boost &quot;Hg.&quot;</th>
<th>F.T. Height - Ft. (without ram)</th>
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<tbody>
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<td>Main</td>
<td>Take-Off</td>
<td>2000</td>
<td>2700</td>
<td>54</td>
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<tr>
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<td>1650</td>
<td>2700</td>
<td>52.5</td>
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Section II - Performance

Army Air Forces Materiel Centre Memorandum Report on

Fighter Single Engine F4U-1, Serial No. 02296

March 3, 1943

Subject: Flight Tests
Section: Flight
Serial No.: FS-M-19-1554-A

A. Purpose.

1. Report on performance tests of Vought-Sikorsky F4U-1 airplane at Wright Field. Airplane equipped with Pratt & Whitney two-stage R-2800-8 engine and a three-bladed constant speed propeller, blade design No. 6443A-21. Gross weight at take-off was 11,936 pounds. Landing gear retracted; wing flaps neutral; cowl, oil, and intercooler flaps neutral; carburetor cold; mixture auto-rich; six .50 caliber wing guns installed with ports covered with tape. Radio and antenna in place. Horsepowers obtained from power curve R-2800-8, -10, dated 8-25-41.

B. Test Results.

1. High speed in level flight at wide open throttle at 24,750 feet was 388.5 MPH at 2700 RPM with 1525 b.h.p. at 51.5 inches Hg. manifold pressure (Auxiliary stage high ratio).

2. Determination of airspeed indicator and altimeter installation errors.

<table>
<thead>
<tr>
<th>Indicated Airspeed MPH</th>
<th>Indicator vs. Water Column MPH</th>
<th>Calibrated Airspeed MPH</th>
<th>Airspeed Error MPH</th>
<th>Altimeter Installation Error ft At Sea Level</th>
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<tr>
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<td>175</td>
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<td>180</td>
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</table>

Prepared by Nathan R. RosenGarten, 1st Lt. A.C.

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Chief, Flight Section